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By Karl Wieghardt

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I. SEVERAL VORTEX FILAMENTS

In the airfoil theory of Prandtl (reference 1) the wing is replaced by a lifting vortex filament whose circulation varies over the span. By this method the "first problem of airfoil theory," namely, for a given lift distribution to determine the shape of the airfoil, was solved. The inverse "second problem," namely, for a given wing to determine the lift distribution, was then solved by Betz (reference 2), the computation being simpler for small aspect ratios than for large ones. For the latter, an approximate solution was obtained by Trefftz (reference 3). The answer was thus found to the most important practical question, namely, the manner in which the wing forces are distributed along the span.

The chordwise distribution theory was simply taken over from the theory of the infinite wing. The Ackermann formulas, published by Birnbaum (reference 4), in which the infinite wing was replaced by a plane vortex sheet — on account of their linearized form permit also application to the finite wing and this application was carried out by Blenk (reference 5) for the rectangular wing. Since in this work a series expansion in b/t was used, the computation converges only for large aspect ratios. In the present paper a useful approximate solution will be found also for wings with large chord - i.e., small aspect ratio.

Another method of investigating the lift distribution along the two dimensions (span and chord) was found by Prandtl (reference 6) in his use of the acceleration potential. This method assumes, however, that the potential is known for a suitable number of source distribu-

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tions over the horizontal projection of the wing. The first application was made by Kinner (reference 7) in his work on the wing with circular plan form, since these functions are obtainable for the circle, The method appears, however, for the present to offer no promise for the rectangular wing, since no expansion of the potential into a series of known functions is known for the rectangle. Bar this reason the computation In the present paper will still be conducted by the vortex-shset method.

For accurate investigation of the lift distribution, the wing nust be replaced by a vortex sheet. A qood idea of the distribution can still be obtained if the wing is represented by a finite number of discrete vortex filaments, and the necessary amount of computation is there-by reduced considerably as compared with the continuous circulation distribution. This is because in the case of the vortex sheet, the condition that the component of the induced velocity at right angles to the wing should be equal to that due to the flow, gives rise to an integral equation. Bor individual vortex filaments, however, this flow condition need be satisfied exactly only at single points, so that only a system of linear equations is obtained. Figure 1 shows such a vortex system, for which the computation was carried out. In order that the results obtained from using only a few vortices, or even a single one. be as accurate as possible, the distance of the first vortex filament from the leading edge is taken to be $a/4$. It is known from previous work that the circulation in the neighborhood of the leading edge increases as $1/\sqrt{x}$; the foremost, strongest vortex aiich gives the circulation contribution from tho leading edge to the foremost points considered, then lies exactly at the center of pressure of the forward lift portion because the center of gravity of $y = c/\sqrt{x}$ lieu at $s = x/3$. The points at which the total velocity at right angles to tho wing is made to vanish, lie in the center between two vortex lines and at $x = t - a/4$. Tho wing is a plane rectangular plnto of zoro thickness with chord $t = na$ for n vortices. Tho notation is indfcated in figure 1. The coordinates of_tho point A aro x^* and y^*. The velocity at right angles to the xy plane. induce1 by the bound and trailing vortices at the point A, is then given by the Biot-Savart law:

$$
\left.
\begin{aligned}
w_A = \sum_1^n i \Bigg\{ & \frac{\bar{x}_i}{4\pi} \int_{-b/2}^{+b/2} \frac{\Gamma_i(y)\, dy}{[\bar{x}_i^2 + (y - y^*)^2]^{3/2}} \\
& - \frac{\bar{x}_i}{4\pi} \int_{-b/2}^{+b/2} \frac{\frac{d\Gamma_i(y)}{dy}}{\sqrt{\bar{x}_i^2 + (y - y^*)^2}} \frac{dy}{y - y^*} \\
& - \frac{1}{4\pi} \int_{-b/2}^{+b/2} \frac{1}{y - y^*} \frac{d\Gamma(y)}{dy}\, dy \Bigg\}; \quad \bar{x}_i = x^* - x_i
\end{aligned}
\right\} \quad (1)
$$

The first integral, which arises from the bound vortices, gives after integration by parts:

$$
\left[\frac{(y - y^*)\, \Gamma_i(y)}{\bar{x}_i^2 \sqrt{\bar{x}_i^2 + (y - y^*)^2}} \right]_{-b/2}^{+b/2}
$$

$$
- \int_{-b/2}^{+b/2} \frac{(y - y^*)}{\bar{x}_i^2 \sqrt{\bar{x}_i^2 + (y - y^*)^2}} \frac{d\Gamma_i(y)}{dy}\, dy
$$

The first expression vanishes since the circulation at the tips must be zero; $\Gamma\left(\pm \frac{b}{2}\right) = 0$. There is thus obtained w_A:

$$
w_A = -\frac{1}{4\pi} \sum_1^n i \int_{-b/2}^{+b/2} \frac{d\Gamma_i(y)}{dy} \frac{1}{y - y^*} \left(1 + \frac{\sqrt{\bar{x}_i^2 + (y - y^*)^2}}{\bar{x}_i}\right) dy \qquad (1a)
$$

Since $\Gamma_i(y)$ decreases from the center of the plate toward the tips, $\frac{d\Gamma_i(y)}{dy} < 0$, and hence $w_A > 0$. From the condition $w_A = V \sin \alpha$, there are obtained for m points m equations $w_{A_1} = w_{A_2} = \ldots = w_{A_m} = V \sin \alpha$. It is thus possible either to assume the same spanwise circulation distribution - for example, the elliptic for all n = m vortices - or set up a series expansion with r undetermined coefficients for $n = m/r$ vortices.

An example by the second method will first be computed. For this purpose, the following transformation of coordinates is made:

$$y = \frac{b}{2}\cos\varphi, \quad y^* = \frac{b}{2}\cos\varphi^*; \qquad \bar{x}_i = x^* - x_i = \frac{b}{2}\delta_i$$

so that $-\frac{b}{2} \leq y \leq \frac{b}{2}$ corresponds to $\pi \geq \varphi \geq 0$. Equation (1a) then becomes:

$$w_A = \frac{1}{2\pi b}\sum_1^n i \int_0^\pi \frac{1}{\cos\varphi - \cos\varphi^*} \left(1 + \frac{\sqrt{\delta_i^2 + (\cos\varphi - \cos\varphi^*)^2}}{\delta_i}\right)\frac{d\Gamma_i(\varphi)}{d\varphi}\,d\varphi \qquad (2)$$

For each $\Gamma_i(\varphi)$, a trigonometric series that contains only the $\sin(2\nu + 1)\varphi$ terms was assumed, since the relations are assumed symmetric with respect to the wing center: $\Gamma_i(\varphi) = \Gamma_i \sin\varphi\,(1 + a_i^{(1)}\sin\varphi + a_i^{(2)}\sin 3\varphi)$. Thus for each vortex filament, there are three undetermined coefficients Γ_i, $a_i^{(1)}$, and $a_i^{(2)}$. Be then have

$$\frac{d\Gamma_i(\varphi)}{d\varphi} = \Gamma_i(\cos\varphi + 2a_i^{(1)}\sin\varphi\cos\varphi + a_i^{(2)}\cos\varphi\sin 3\varphi + 3a_i^{(2)}\sin\varphi\cos 3\varphi)$$

Substituting this expression in equation (2), there is obtained:

$$w_A = -\frac{1}{2\pi b}\sum_1^n i\,\Gamma_i\left\{\frac{1}{8\delta_i}J - \pi + a_i^{(1)}\,f_1(\delta_i,\cos\varphi^*)\right.$$

$$\left. + a_i^{(2)}\,f_2(\delta_i,\cos\varphi^*)\right\} \qquad (3)$$

where the functions f_1 and f_2 are made up of integrals which may be evaluated by elementary methods. The integral J, which is also a function of δ_i and $\cos\varphi^*$ is, in any particular case to be determined by graphical or numerical methods. The integral is

$$J = \int_0^\pi \frac{\cos\varphi\ d\varphi}{\cos\varphi - \cos\varphi^*}\sqrt{\delta_i^2 + (\cos\varphi - \cos\varphi^*)^2} =$$

$$= \int_0^\pi \frac{(\cos\varphi - \cos\varphi^*)}{\delta_i^2 + \sqrt{\delta_i^2 + (\cos\varphi - \cos\varphi^*)^2}}\cos\varphi\ d\varphi + \pi\,\delta_i$$

for δ_i and $\cos^* > 0$. Through this transformation the singularity at $\varphi \rightarrow \varphi^*$ has bean removed. Since for each vortex line there are three undetermined coefficients, the flow condition can be satisfied for each set of three points between two lines and for three points at x = t - a/4. On account of tie symmetry 3n different points may be chosen on a half wing and for the corresponding points, symmetrical with respect to the center line of the plate, the condition WA = V sin a is then automatically satisfied. Altogether, therefore, the flow condition is accuratelp satisfied for 6n points or, in case one of each set of three points lies on the center line, for 5n points. The entire computation is based on the expectation that the condition $w_A = V\sin\alpha$ will be, on the average, satisfied at least approximately, also at other points of the surface, and that the singular behavior of w_A along each of the lifting vortices will not have too great an effect on the approximate computation of the circulati although aerodynamically this can only be justified':; considering the plate as replaced by several wings lying one behind the-other, each represented by a vortex filament. The choice of the number of vortex filaments n is, for practical reasons, restricted since while the number of points considered increases only linearly with n, the required computation work of solving the system of 3n equations increases at a greater rate.

The numerical computation was carried out for the following case: $n = 4$, $\varphi^* = 30^\circ$ (150°), 60° (120°), and 90° (center line) with $b = 4a$, corresponding to an aspect ratio of $A = b/t = 1$. (See fig. 2.) The integral J ($J(+\delta_i) = J(-\delta_i)$) was determined for the four values 1/4, 3/4, 5/4, 7/4 which are assumed by δ_i and, on account of the symmetry, for only three values of cos φ^*. For $\varphi^* = 90^\circ$, an elliptic integral of the second kind is obtained for J. From the functions f_1 (δ_i, cos φ^*) and f_2 (δ_i, cos φ^*), the coefficients were obtained for a system of 12 equations which was solved by the usual elimination process with the computation machine since the system could not be solved by iteration. As the computation was carried out to only five decimal places, it was afterwards found to be of insufficient accuracy for the determination of the last three unknowns; the circulation of the rearmost vortex filament, therefore, could only be estimated by extrapolation. For the remaining circulations, there was obtained:

$$\Gamma_1 = +0.737_0 \, bV \sin \alpha \qquad \Gamma_2 = +0.116_2 bV \sin \alpha \qquad \Gamma_3 = +0.058_{43} bV \sin \alpha$$

$$a_1^{(1)} = -0.136_3 \qquad a_2^{(1)} = +0.564_2 \qquad a_3^{(1)} = +0.245_5$$

$$a_1^{(2)} = +0.005_7 \qquad a_2^{(2)} = +0.001_2 \qquad a_3^{(2)} = -0.061_1$$

Those circulation distributions are shown on figure 3.

Integrating over the span:

$$\rho \, V \frac{b}{2} \int_0^\pi \Gamma_i(\varphi) \sin \varphi \, d\varphi = \rho \, V \frac{b}{2} \Gamma_i \left(\frac{\pi}{2} + \frac{4}{3} a_i^{(1)} - \frac{4}{15} a_i^{(2)} \right)$$

these may be considered by the Kutta-Joukowsky theorem as the lift contributions of the individual wing strips (along the chord), The lift is then distributed as follows:

From the leading edge to 3t/16: A_I -0.512, $\rho b t V^2 \sin \alpha$

From 3t/16 -to 7t/16: $A_{II} = 0.133_5$ "

From 7t/16 to 11t/16: $A_{III} = 0.055s$ "

From 11t/16 to trailing edge $A_{IV} \approx 0.02$ "

The total lift is $A = \Sigma_\mu A_\mu = 0.72s\ \rho\ b\ t\ V^2 \sin \alpha$ and the lift coefficient $c_a = A / \frac{\rho}{2} b\ t\ V^2 = 1.44, \sin \alpha$. The moment about the leading edge is obtained as the sum of the moments of the several strips: $M = t/16\ (A_I + 5A_{II} + 9A_{III} + 13A_{IV})$ ond the moment coefficient $c_m = 0.24_3 \sin a$. The position of the center of pressure is' obtained from $s = \frac{c_m}{c_a} t = 0.16, t$, where s is the distance of the center of pressure from the leading edge. Finally, for the forward three vortices the factor $v_i = \frac{\int_0^\pi \Gamma_i(\varphi) \sin \varphi\, d\varphi}{2(1+a_i^{(1)}-a_i^{(2)})}$ can be determined: $v_1 = 0.80_9$, $v_2 = 0.74,$. $v_3 = 0.73,$.

Since for all vortices this factor is approximately 'equal to $\pi/4 = 0.78_{54}$, it appears justifiable - at least, for deep wings, that is, small aspect ratios A = b/2 - to assume initially an elliptic spanwise lift distribution and so considerably simplify the computation. The fact that this assumption. according to the above computed example, is not quite applicable to the roar vortices, is of no great importance on account of the strong rearward drop in the circulation,

Since for elliptic distribution there is only one undetormined coefficient for each vortex line,. namely, the circulation Γ_i in the center of the span, the flow condition $w_A = V \sin \alpha$ can also be satisfied at only one point (and the point symmetrical with resgnct to the center line) between each two lines and at $x = 15t/16$, For this reason, the assumed points are taken on tie center line and in the center between each two succeeding lines, and the last at $x = 15t/16$. We then have $y^* = 0$, $x^* = \frac{1}{2}(x_k+x_{k+1})$, for the rearmost point $x^* = 15t/16$ and $\bar{x}_1 = \pm \frac{t}{2n}, \pm \frac{at}{2n}, \ldots + \frac{2(n-1)t}{2n}$, where n is the number of vortices. Substituting $\Gamma_1(y) = \Gamma_1 \sqrt{1 - \left(\frac{y}{b/2}\right)^2}$, $\frac{d\Gamma_1(y)}{dy} = -\frac{\Gamma_1 y}{\left(\frac{b}{2}\right)^2 \sqrt{1 - \left(\frac{y}{(b/2)}\right)^2}}$

into equation (1a), there is obtained for the induced velocity at the point considered:

$$w_A = \frac{1}{\pi b^2} \sum_1^n i\, \Gamma_1 \int_{-b/2}^{+b/2} \frac{1}{\sqrt{1 - \left(\frac{y}{b/2}\right)^2}} \left(1 + \frac{\sqrt{\bar{x}_1^2 + y^2}}{\bar{x}_1}\right) dy$$

$$= \frac{1}{\pi b^2} \sum_1^n i\, \Gamma_1 \left\{\pi \frac{b}{2} + \frac{2}{\bar{x}_1} \int_{-b/2}^{+b/2} \sqrt{\frac{\bar{x}_1^2 + y^2}{1 - \left(\frac{y}{b/2}\right)^2}}\, dy\right\}$$

The elliptic second integral is reduced to the normal form by the substitution $y = b/2 \cos\psi$, so that there is obtained:

$$w_A = \frac{1}{\pi b} \sum_1^n i\, \Gamma_1 \left\{\frac{\pi}{2} + \frac{\sqrt{\bar{x}_1^2 + (b/2)^2}}{\bar{x}_1}\, E\left(\frac{b/2}{\sqrt{\bar{x}_1^2 + (b/2)^2}}, \frac{\pi}{2}\right)\right\} \tag{4}$$

where E is the complete elliptic integral of the second kind $E\left(k, \frac{\pi}{2}\right) = \int_0^{\pi/2} \sqrt{1 - k^2 \sin^2\psi}\, d\psi$ with modulus $k = \frac{b/2}{\sqrt{\bar{x}_1^2 + (b/2)^2}}$. This function is tabulated, for example, in Jahnke-Emde: Funktionentafeln. It has the following limiting value for $k = \sin\varphi$, $0 \leq k \leq 1$, $0 \leq \varphi \leq \pi/2$, and in the above equation for $\infty \geq |\bar{x}_1| \geq 0$: $\frac{\pi}{2} \geq E\left(\varphi, \frac{\pi}{2}\right) \geq 1$.

The expression in parentheses in equation (4) gives the coefficients for the linear nonhomogeneous system of equations for the n unknowns $\Gamma_1, \Gamma_2, \ldots \Gamma_n$. Putting 2b on the right side, it reads $2b\,V \sin\alpha$ for all equations, on account of the condition $w_A = V \sin a$. The coefficients of the principal diagonal are all equel and similarly in eaoh diagonal from upper left to lower right. Thus, for example, for $\lambda = 6$, $n = 4$; that is, $x_1 = \pm b/40$, $\pm 3b/48$, $\pm 5b/48$, $\pm 7b/48$, the system becomes:

$$16.346_1\,\Gamma_1 - 14.346_1\,\Gamma_2 - 4.250_5\,\Gamma_3 - 2.284_2\,\Gamma_4 = 2b\,V \sin a$$

$$6.250_5\,\Gamma_1 + 16.346_1\,\Gamma_2 - 14.346_1\,\Gamma_3 - 4.250_5\,\Gamma_4 = 2b\,V \sin\alpha$$

$$4.284_2\,\Gamma_1 + 6.250_5\,\Gamma_2 + 16.346_1\,\Gamma_3 - 14.346_1\,\Gamma_4 = 2b\,V \sin\alpha$$

$$3.470_4\,\Gamma_1 + 4.284_2\,\Gamma_2 + 6.250_5\,\Gamma_3 + 16.346_1\,\Gamma_4 = 2b\,V \sin\alpha$$

with the solutions

$$\Gamma_1 = 1.3344\,t\,V \sin\alpha \qquad \Gamma_3 = 0.328_0\,t\,V \sin\alpha$$

$$\Gamma_2 = 0.558_3\,t\,V \sin\alpha \qquad \Gamma_4 = 0.179_1\,t\,V \sin\alpha$$

The lift contributions from the four strips of the wing are: $A_I = 1.048_1\,b\,t\,V \sin a$, $A_{II} = 0.478_5\,b\,t\,V \sin a$, $A_{III} = 0.257_5\,b\,t\,V \sin a$, $A_{IV} = 0.143_1\,b\,t\,V \sin\alpha$; the coefficients: $c_a = 3.76_5 \sin\alpha$, $c_w = 0.754 \sin^2 a$, $c_m = 0.92_3 \sin a$, and the distance of the center of pressure from tho leading edge $s = 0.24_1\,t$. A comparison of the results by this method and those by the Vortex-surface method will be made in the third section.

Still simpler is the limiting case of the above systen for $n = 1$. The conputation is then made with a vortex with spanwise elliptic circulation distribution at the diatanco $t/4$ from the leading edge. Tho single point consfdorod lies at $x = 3/4\,t$. The center of pressure, according to the assumption thon, always lies at $s = t/4$, which is sufficiently accurate for rather large aspect ratios $\lambda > 3$, as shown by the conputation with tho vortex shoet. The circulation and the coofficients in this case can, in general, be explicitly expressed:

$$\Gamma = \frac{2b\,V\,\sin\alpha}{1 + \frac{2}{\pi}\sqrt{1+\lambda^2}\;E\left(\frac{\lambda}{\sqrt{1+\lambda^2}},\frac{\pi}{2}\right)};$$

$$c_a = \frac{\pi\,\lambda\,\sin\alpha}{1 + \frac{2}{\pi}\sqrt{1+\lambda^2}\;E\left(\frac{\lambda}{\sqrt{1+\lambda^2}},\frac{\pi}{2}\right)},\quad c_{wi} = \frac{1}{\pi\lambda}\,c_a^2,\quad c_m = \frac{1}{4}\,c_a$$

With infinitely increasing aspect ratio there is thus obtained a limiting value for c_a: $\lim_{\lambda\to\infty} c_a = \frac{\pi^2}{2}\sin\alpha$. From the potential theory, on the other hand, there is obtained for the wing with infinite span $c_a = 2\pi\sin\alpha$. This difference is readily explainable from the fact that elliptic circulation distribution (that is, factor 7, previously defined, equals $\pi/4$) has been assumed above. According to Betz, however, this factor for rectangular wings becomes larger with increasing λ up to the limiting value 1 for $\lambda\to\infty$. Hence, multiplying by $4/\pi$, we have actually $\frac{4}{\pi}\lim_{\lambda\to\infty} c_a = (c_a)_{pot}$.

It is now simple to refine this method by assuming a trigonometric series for $\Gamma(y)$: $y = \frac{b}{2}\cos\varphi$, $\Gamma(\varphi) = \Gamma\sin\varphi\,(1 + a_1\sin\varphi + \ldots a_n\sin n\varphi)$. Then, again, equation (3) is obtained – this time for one vortex with $\delta_1 = 1/\lambda$; that is, without the summation sign. The drag is determined

from $W = \rho\,\frac{b}{2}\int_0^\pi \Gamma(\varphi)\,w(\varphi)\sin\varphi\,d\varphi$, where $w(\varphi) = \frac{1}{2\pi b}\int_0^\pi \frac{1}{\cos\varphi - \cos\varphi_t}\frac{d\Gamma(\varphi)}{d\varphi}\,d\varphi$, and there is obtained $c_{wi} = \left(\frac{\pi}{2} + \frac{4}{3}a_1 + \frac{2}{\pi}a_1^2 + \ldots\right)\frac{\Gamma^2}{V^2 bt}$. In the examples there was set: $\Gamma(\varphi) = \Gamma\sin\varphi\,(1 + a_1\sin\varphi)$, $\varphi_1^* = 30^\circ$, $\varphi_2^* = 60^\circ$, which corresponds to $y_1^* = \frac{\sqrt{3}}{4}b$, $y_2^* = \frac{b}{4}$. On figure 4 the lift-distribution coefficient is plotted against the aspect-ratio. It may be seen that the experimentally obtained lift coefficients (reference 8) are very closely approached by this simple computation.

The following table is to be used in connection with figure 4:

λ	c_a	c_w	c_m	v
$\to 0$	$\to \frac{\pi}{2} \lambda \sin \alpha$	$\to \frac{\pi}{4} \lambda \sin^2 \alpha$	$\to \frac{\pi}{8} \lambda \sin \alpha$	$\to \frac{\pi}{4}$
4/7	$0.89_6 \sin \alpha$	$0.41_7 \sin^2 \alpha$	$0.22_4 \sin \alpha$	0.785_7
4/3	$1.81_4 \sin \alpha$	$0.78_5 \sin^2 \alpha$	$0.45_4 \sin \alpha$	0.788_9
6	$4.202_2 \sin \alpha$	$0.93_2 \sin^2 \alpha$	$1.05_1 \sin \alpha$	0.838_7

II. VORTEX SHEET

With the method of discrete vortex filaments, a further refinement in the lift distribution - hence an increase in the accuracy by increasing the number of vortices - is practically excluded on account of the computation labor involved. It was therefore carried to the limit $n \to \infty$, and an attempt was made by analytical methods to restrict the computation work as far as possible. On increasing n, the circulation contributed by each of the vortices and their distances apart become smaller up to the limiting case $n \to \infty$, when the circulation distribution becomes a surface distribution. The dimensions of this surface distribution $\gamma(x,y)$ are circulation per unit chord, that is, centimeters per second. An infinitesimal vortex $\gamma(x,y)\,dx$ induces, according to equation (1), at any point of the surface, the velocity:

$$dw_A = \frac{1}{4\pi} \int_{-b/2}^{+b/2} \frac{(x^*-x)\ \gamma(x,y)\ dx}{[(x^*-x)^2 + (y-y^*)^2]^{3/2}}\, dy$$

$$= \frac{1}{4\pi} \int_{-b/2}^{+b/2} \frac{1}{y-y^*} \frac{\partial \gamma(x,y)}{\partial x}\, dx \left[1 + \frac{x^* - x}{\sqrt{(x^*-x)^2 + (y-y^*)^2}}\right] dy$$

where α is assumed small enough so that $\sin^2 \alpha \approx 0$ and $\cos \alpha \approx 1$. The equation would be strictly true if the trailing vortices were sbod in the direction of the wing. Integrating the first integral by parts, there is again obtained, on account of $\gamma(x, y = \pm b/2) = 0$,

$$w_A = -\frac{1}{4\pi}\int_0^t dx \int_{-b/2}^{+b/2} \left\{\frac{\sqrt{(x^*-x)^2+(y-y^*)^2}}{(x^*-x)\ (y-y^*)} + \frac{1}{y-y^*}\right\} \frac{\partial\gamma(x,y)}{\partial y}\, dy$$

The condition $w_A = V \sin \alpha$ gives, after a transformation of coordinates,

$$\begin{aligned} x &= t/2(1+\xi) \\ y &= b/2\ \eta \end{aligned} \qquad \int_{-1}^{+1}\int_{-1}^{+1} \left\{\frac{\sqrt{(\xi^*-\xi)^2+\lambda^2(\eta-\eta^*)^2}}{(\xi^*-\xi)\ (\eta-\eta^*)}\right.$$

$$\left. + \frac{1}{\eta-\eta^*}\right\} \frac{\partial\gamma(\xi,\eta)}{\partial\eta}\, d\xi\, d\eta = 4\pi\, A\, V \sin \alpha \qquad (5)$$

This is a two-dimensional linear integral equation of the first kind in $\frac{\partial\gamma(\xi,\eta)}{\partial\eta}$. On account of the singularity of the kernel at the point considered for $\begin{matrix}\xi \to \xi^* \\ \eta \to \eta^*\end{matrix}$, this equation could not be solved even approximately since the usual method of approximating the kernel by a polynomial, assumes continuity. Also the particular property of the kernel on which it depends $(\xi - \xi^*)$ and $(\eta - \eta^*)$ could not be uaed for a solution method.

The only recourse therefore is to simplify the integral equation by au aerodynamically reasonable assumption. It was thus assumed that the spanwise circulation distribution is the same for all ξ. This results in a lowering of the order of the integral equation. although the flow condition can then no longer be satisfied over the entire wing but only on a straight line η^* = const. For obvious reasons elliptic distribution was assumed over the span. and $\eta^* = 0$; that le. the flow condition was to be satisfied on the center line of the plate. For these points of the surface then, the condition $v_A = V \sin \alpha$ is required, and for other point's it is expected that the deviation fron this condition-la not too large.

With $\gamma(\xi,\eta) = \gamma(\xi)\sqrt{1-\eta^2}$, $\frac{\partial\gamma(\xi,\eta)}{\partial\eta} = -\frac{\eta}{\sqrt{1-\eta^2}}\gamma(\xi)$

and $\eta^* = 0$, equation (5) simplifies to the following:

$$\int_{-1}^{+1}\gamma(\xi)d\xi\int_{-1}^{+1}\left\{\frac{\sqrt{(\xi^*-\xi)^2+\lambda^2\eta^2}}{(\xi^*-\xi)\eta}+\frac{1}{\eta}\right\}\frac{\eta}{\sqrt{1-\eta^2}}\,d\eta = 4\pi\lambda\ V\ \sin\alpha$$

In the integration with respect to η, on elliptic integral is again obtained:

$$\int_{-1}^{+1}\frac{\sqrt{(\xi^*-\xi)^2+\lambda^2\eta^2}}{(\xi^*-\xi)\sqrt{1-\eta^2}}\,d\eta =$$

$$2\sqrt{1+\frac{\lambda^2}{(\xi^*-\xi)^2}}\int_0^{\pi/2}\sqrt{1-\frac{\lambda^2}{\lambda^2+(\xi^*-\xi)^2}\sin^2\varphi}\;d\varphi,$$

with $\eta = \cos\varphi$, so that

$$\int_{-1}^{+1}\left\{\frac{\sqrt{(\xi^*-\xi)^2+\lambda^2}}{(\xi^*-\xi)}\,E\left(\frac{\lambda}{\sqrt{\lambda^2+(\xi^*-\xi)^2}},\frac{\pi}{2}\right)+\frac{\pi}{2}\right\}\gamma(\xi)\,d\xi = 2\pi\lambda\ V\ \sin\alpha^{**} \qquad (6)$$

** Corresponding integral equations can also be set up for. airfoils with arbitrary plan form symmetrical with respect to the center line. If $b(\xi)$ la the span, varying with the chord, equation (5) becomes:

$$\lambda(\xi) = \frac{b(\xi)}{t}: \int_{-1}^{+1}\int_{-1}^{+1}\left\{\frac{\sqrt{(\xi^*-\xi)^2+\lambda^2(\xi)(\eta-\eta^*)^2}}{(\xi^*-\xi)(\eta-\eta^*)}\right.$$

$$\left.+\frac{1}{\eta-\eta^*}\right\}\frac{\partial\gamma(\xi,\eta)}{\partial\eta}\frac{1}{\lambda(\xi)}\,d\xi\,d\eta = 4\pi\ V\ \sin a$$

(Continued on p. 14)

Equation (5) is thus reduced to a one-dimensional integral equation. The kernel, it is true, has become more complicated, and the singularity for $\xi \to \xi^*$ naturally, still remains. For the approximate solution of this equation, there is assumed for $\gamma(\xi)$ a series of functions which Birnbaum has used in his paper

$$\gamma(\xi) = A_1 \sqrt{\frac{1-\xi}{1+\xi}} + A_2 \sqrt{1-\xi^2} + A_3 \xi \sqrt{1-\xi^2} + A_4 \xi^2 \sqrt{1-\xi^2} \qquad (7)$$

The four undetermined coefficients A_1, A_2, A_3, A_4 are so determined that the integral equation is satisfied at four points ξ_1 to ξ_4. Since the kernel contains λ, it is necessary to compute each time for a definite aspect ratio. The integrations, on account of the complicated kernel, must be carried out graphically or numerically. The procedure of the very laborious computation thus, is the following:

The four basic functions $\gamma_1, \gamma_2, \gamma_3$, and γ_4 for a series of values of ξ from -1 to +1, are first computed; then for the same arguments for a definite λ, the kernel functions $k(\lambda;\xi^*,\xi)$ for the four points considered. (In the examples carried out $\xi_1^* = -\frac{1}{2}, \xi_2^* = 0$, $\xi_3^* = +\frac{1}{2}, \xi_4^* = +1$; then, on account of $k(-\xi^*,-\xi) = k(\xi^*,\xi) + \pi$, $k(-0.5,\xi) = k(+0.5,\xi) + \pi$.) Each of these kernel functions is multiplied by $\gamma_1(\xi), \gamma_2(\xi), \gamma_3(\xi)$, $\gamma_4(\xi)$:

**(Continued from p. 13)

For the elliptic wing there is obtained for elliptic circulation distribution over the span with $\lambda(\xi) = b_{max}/t \sqrt{1-\xi^2}$:

$$\int_{-1}^{+1} \left\{ \frac{\sqrt{(\xi^* - \xi)^2 + \lambda^2 (1-\xi)^2}}{(\xi^* - \xi)} E \left(\frac{\lambda \sqrt{1-\xi^2}}{\sqrt{\lambda^2 (1-\xi^2) + (\xi^* - \xi)^2}}, \frac{\pi}{2} \right) + \frac{\pi}{2} \right\} \frac{\gamma(\xi)}{\lambda \sqrt{1-\xi^2}} d\xi = 2\pi V \sin \alpha; \quad \lambda = \frac{b_{max}}{t}$$

This equation is solved approximately as in the case of the rectangular wing. The limiting case $\lambda \to \infty$ gives for $\frac{1}{2\pi} \int_{-1}^{+1} \frac{1}{\xi^* - \xi} \gamma(\xi) d\xi = V \sin \alpha$ the solution of the potential theory $\gamma(\xi) = 2V \sin \alpha \sqrt{\frac{1-\xi}{1+\xi}}$ with $c_a = 2\pi \sin \alpha$, $c_m = \frac{\pi}{2} \sin \alpha$, $s = \frac{1}{4} t$.

$$i_1(\xi) = k(\xi_1^*,\xi)\,\gamma_1(\xi) \qquad i_5(\xi) = k(\xi_2^*,\xi)\,\gamma_1(\xi)$$

$$i_2(\xi) = k(\xi_1^*,\xi)\,\gamma_2(\xi) \qquad i_6(\xi) = k(\xi_2^*,\xi)\,\gamma_2(\xi)$$

$$i_3(\xi) = k(\xi_1^*,\xi)\,\gamma_3(\xi)$$

$$i_4(\xi) = k(\xi_1^*,\xi)\,\gamma_4(\xi)$$

etc. to

$$i_{16}(\xi) = k(\xi_4^*,\xi)\,\gamma_4(\xi)$$

Those 16 functions must now be integrated graphically or numerically: for example, by the Simpson rule:

$$I_k = \int_{-1}^{+1} i_k(\xi)\,d\xi$$

Since the $k(\xi^*,\xi)$ and hence the $i(\xi)$ become infinite for $\xi \rightarrow \xi^*$, it is necessary in the quadrature to exclude a region $\xi^* - \epsilon < \xi < \xi^* + \epsilon$ and approximately determine by analytical methods the Cauchy principal value for $\int_{\xi-\epsilon}^{\xi+\epsilon} i(\xi)\,d\xi$. We thus finally have the coefficients I_k for the linear nonhomogeneous system of equations:

$$\sum_{\nu=1}^{4} I_{4\mu+\nu}\,A_\nu = 2\pi\,\lambda\,V \sin\alpha \qquad \text{for } \mu = 0, 1, 2, 3, \ldots$$

from which the $A_\nu(\lambda)$ can be determined. We then have:

$$\gamma(\lambda;\xi,\eta) = \sum_1^4 \nu\,A_\nu(\lambda)\,\gamma_\nu(\xi)\sqrt{1-\eta^2}.$$ In this manner the circulation distribution was determined for the aspect ratios $A = 1/4$, $1/2$, and 1.

For greater aspect ratios, $A > 2$, the 16 graphical quadratures are not required, but the approximate computation can be carried out analytically. For this purpose the elliptic integral must approximately evaluated for $45^\circ < \varphi \leq 90^\circ$. Since $\sin\varphi = \dfrac{\lambda}{\sqrt{\lambda^2+(\xi^*-\xi)}}$ and $|\xi^*-\xi| \leq 2$,

$\varphi = 45^\circ$ is tha smallest argument for $\lambda \geq 2$. For this range $E(x, \pi/2)$ was replaced by $\overline{E}(x, \pi/2) = 1 + 0.44\sqrt{1 - x^2}$. At first there was set $\overline{E}(x, \pi/2) = 1 + \left(\frac{\pi}{2} - 1\right)\sqrt{1 - x^2}\,\sum_0^n \nu a_\nu x^\nu$, but the computation then became so complicated that there was no advantage wined over the previous method; $\overline{E}$ is thus determined with an error which is approximately -3 percent for $\varphi = 45^\circ$, +3.6 percent (maximum value) for $\varphi = 80^\circ$, and approaches zero as $\varphi \to 90^\circ$. For $\varphi = 90^\circ$, the first derivatives also agree: $\lim\limits_{x \to 1} \dfrac{d\overline{E}\left(x, \frac{\pi}{2}\right)}{dx} \to -\infty$ and

$$\lim_{x \to 1} \frac{dE\left(x, \frac{\pi}{2}\right)}{dx} = \lim_{x \to 1} \int_0^{\pi/2} \frac{-x \sin\varphi}{\sqrt{1 - P \sin^2\varphi}}\, d\varphi \to -\infty$$

For $\lambda > 2$ the radical can be developed into a power series which converges for ell ξ^*, since $|\xi^* - \xi| \leq 2$. With

$$E\left(\frac{\lambda}{\sqrt{\lambda^2 + (\xi^* - \xi)^2}}, \frac{\pi}{2}\right) \approx \overline{E}\left(\frac{\lambda}{\sqrt{\lambda^2 + (\xi^* - \xi)^2}}, \frac{\pi}{2}\right) =$$

and

$$1 + 0.44\,\frac{|\xi^* - \xi|}{\sqrt{\lambda^2 + (\xi^* - \xi)^2}}$$

and

$$\sqrt{\lambda^2 + (\xi^* - \xi)^2} \approx \lambda\left[1 + \frac{1}{2}\left(\frac{\xi^* - \xi}{\lambda}\right)^2 - \frac{1}{8}\left(\frac{\xi^* - \xi}{\lambda}\right)^4 + \frac{1}{16}\left(\frac{\xi^* - \xi}{\lambda}\right)^6\right]$$

equation (6) then becomes:

$$\int_{-1}^{+1}\left\{\frac{\lambda}{\xi^* - \xi} + \frac{1}{2\lambda}(\xi^* - \xi) - \frac{1}{8\lambda^3}(\xi^* - \xi)^3 + \frac{1}{16\lambda^5}(\xi^* - \xi)^5\right\}\gamma(\xi)\,d\xi$$

$$+ \int_{-1}^{+1}\left(0.44\,\mathrm{sign}(\xi^* - \xi) + \frac{\pi}{2}\right)\gamma(\xi)\,d\xi = 2\pi A v \sin a$$

Again substituting $\gamma(\xi)$ from equation (7) and integrating, there ore obtained on the left side for the coefficients of A_1, A_2, A_3, A_4, the following functions of λ and ξ^*.

$$F_1(\lambda,\xi^*) = \pi\left[\frac{\xi^{*5}}{16\lambda^5} + \frac{\xi^{*4}}{32\lambda^5} + \left(\frac{5}{16\lambda^5} - \frac{1}{8\lambda^3}\right)\xi^{*3}\right.$$

$$+ \left(\frac{15}{64\lambda^5} - \frac{3}{16\lambda^3}\right)\xi^{*2} + \left(\frac{15}{128\lambda^5} - \frac{3}{16\lambda^3} + \frac{1}{2\lambda}\right)\xi^*$$

$$+ \left(\lambda + \frac{1}{4\lambda} - \frac{1}{64\lambda^3} + \frac{5}{256\lambda^5} + \frac{\pi}{2}\right)$$

$$\left. + 0.280_{11}\left(\text{arc sin } \xi^* + \sqrt{1 - \xi^{*2}}\right)\right]$$

$$F_2(\lambda,\xi^*) = \pi\left[\frac{\xi^{*5}}{32\lambda^5} + \left(\frac{5}{64\lambda^5} - \frac{1}{16\lambda^3}\right)\xi^{*3}\right.$$

$$+ \left(\lambda + \frac{1}{4\lambda} - \frac{3}{64\lambda^3} + \frac{5}{256\lambda^5}\right)\xi^* + \frac{\pi}{4}$$

$$\left. + 0.140_{06}\left(\text{arc sin } \xi^* + \xi^*\sqrt{1 - \xi^{*2}}\right)\right]$$

$$F_3(\lambda,\xi^*) = \pi\left[-\frac{5\xi^{*4}}{128\lambda^5} + \left(\lambda + \frac{3}{64\lambda^3} - \frac{5}{128\lambda^5}\right)\xi^{*2}\right.$$

$$\left. - \left(\frac{\lambda}{2} + \frac{1}{16\lambda} - \frac{1}{128\lambda^3} - \frac{5}{2048\lambda^5}\right) - 0.093_{37}\left(1-\xi^{*2}\right)^{3/2}\right]$$

$$F_4(\lambda,\xi^*) = \pi\left[\frac{\xi^{*5}}{128\lambda^5} + \left(\lambda - \frac{3}{64\lambda^3} + \frac{5}{128\lambda^5}\right)\xi^{*3}\right.$$

$$- \left(\frac{\lambda}{2} - \frac{1}{16\lambda} + \frac{1}{128\lambda^3} - \frac{25}{2048\lambda^5}\right)\xi^* + \frac{\pi}{16}$$

$$\left. + 0.035_{01}\left(\text{arc sin } \xi^* - \xi^*\sqrt{1 - \xi^{*2}}\left(1 - 2\xi^{*2}\right)\right)\right]$$

Again a definite value is taken for λ and for four points (here, too, the points chosen were $\xi_1^* = -\frac{1}{2}$, $\xi_2^* = 0$, $\xi_3^* = +\frac{1}{2}$, and $\xi_4^* = +1$) there are computed the 16 coefficient's of the system of equations $\sum_{\nu=1}^{4} F_\nu(\xi_\mu^*)\, A_\nu = 2\pi\lambda V \sin\alpha$ for $\mu = 1, 2, 3, 4$, from which the coefficients A_1, A_2, A_3, A_4 are determined. In this manner the A_ν for $\lambda = 6$ and $\lambda = 2$ are computod. For $\lambda = 2$, this computation. strictly speaking, is at least for $\xi_4{}^* = 1$, not valid because the series into which fho radical was developed is no longer convergent in the limiting case $\xi \rightarrow -1$.

Finally, equation (6) waa'alao considored for tho two linitinq cases $A \rightarrow \infty$ and $A \rightarrow 0$. Bor tho wing of infinito span, $b, \lambda \rightarrow \infty$, $E\left(\frac{\lambda}{\sqrt{\lambda^2+(\xi^*-\xi)^2}}, \frac{\pi}{2}\right)$ becomes oqual to 1 and tho equation goes over into $\int_{-1}^{+1} \frac{1}{\xi^*-\xi}\gamma(\xi)\, d\xi =$ 2π v sin a. Substituting the above expression for $\gamma(\xi)$, there is easily recognized as a solution $\gamma(\xi) = 2V \sin\alpha \sqrt{\frac{1-\xi}{1+\xi}}$, which is the distribution given by the potential theory. Since as $A \rightarrow \infty$, the spanwise distribution becomes $\gamma(\eta) = \text{const}$, this solution satisfies the flow condition at each point of the surface. The same result la also obtained-when in the method of solution for $\lambda > 2$, the $F_\nu(\lambda, \xi^*)$ are considered for very large λ and this system is computed. Then there is also obtained $A_1 = 2$, As $= A_3 =$ A4 $= 0$. The lift coefficient will then be $c_a = \frac{\pi^2}{2}$ sin cc, and the moment coefficient Cm $= \frac{\pi^2}{2}$ sin a ; whereas, according to the two-dimensional potential theory, $c_a = 2\pi$ sin a and $c_m = \frac{\pi}{2}$ sin a. This is again due to the fact that elliptical spanwise distribution was assumed for tho rectangular wing; multiplying, subsequently. by $4/\pi$, the two results become identical. If an elliptical wing is considered and λ is nade to approach infinity, thore is immediatoly obtained $c_a = 2\pi$ sin a and $c_m = \frac{\pi}{4}$ sin a since the roforonce area for tho coefficients is $\pi/4$ b t.

Of considerably greater difficulty is the limiting case of the wing with infinito chord $t \rightarrow \infty$, $\lambda \rightarrow 0$. Here the coordinntoa must be made nondimensional through the span instead of through the chord as heretofore: $x = \frac{b}{2} v$, $x^* = \frac{b}{2} v^*$. Equation (6) with $t \rightarrow \infty$ then goes over into:

$$\int_0^\infty \left\{ \frac{\sqrt{1+(v^*-v)^2}}{v^*-v} \, E\left(\frac{1}{\sqrt{1+(v^*-v)^2}}, \frac{\pi}{2}\right) + \frac{\pi}{2} \right\} \gamma(v)\, dv = 2\pi V \sin\alpha$$

Again it was sought to find for $\gamma(v)$ a series of functions with undetermined coefficients which would then be determined through satisfying the integral equation at several points v^*. In this case, however, no series of functions could be found which at the leading edge $v \rightarrow 0$, increases as $1/\sqrt{v}$ and with $v \rightarrow \infty$, corresponding to the solution for $A = 1/4$, which decreases approximately as $1/v^3$. For this reason, only the following single functions were investigated: $\gamma_1(v) = \frac{A}{\sqrt{v + Bv^3}}$ and $\gamma_2(v) = \frac{C}{\sqrt{v + Dv^5}}$. Since again the integral could be evaluated only graphically or numerically (on account of $0 \leq |v^*-v| < \infty$ a series development of the kernel could not be considered), it was necessary, before substituting, to assume B, and D, respectively, as fixed and then, by quadratures, set up $A(B,v^*)$ and $C(D,v^*)$, respectively, for several values of v^*. In order to limit the integration interval, it was again necessary to make another transformation:

$$u = \frac{1}{1+v}, \qquad u^* = \frac{1}{1+v^*}$$

$$I_1(B;v^*,A) = \int_0^1 \left\{ \frac{\sqrt{1+\left(\frac{u-u^*}{u\,u^*}\right)^2}}{\frac{u-u^*}{u\,u^*}} \, E\left(\frac{1}{\sqrt{1+\left(\frac{u-u^*}{u\,u^*}\right)^2}}, \frac{\pi}{2}\right) + \frac{\pi}{2} \right\} \frac{du}{u^2} \, \frac{A}{\sqrt{\frac{1}{u} - 1 + B\left(\frac{1}{u} - 1\right)^3}}$$

Since the integrand again became singular at two points, namely, at $u \rightarrow u^*$ and $u \rightarrow 1$ ($u = 1$ corresponds to the leading edge, $u = 0$ to the trailing edge), the principal values had to be approximately determined by analytical methods. The entire laborious trial process, however, not with little success as tho effect of the various coefficients on the result was too difficult to estimate. As an approximation, it is possible to sot at nost $\Upsilon(v) \approx -\frac{1.12}{\sqrt{v} + 2.4\, v^3}$; for the range $0.1 \leq v < \infty$, the downwash orror with respect to $v \sin \alpha$ then amounts to about 7 percent - this error, however, strongly increasing toward the leading edge $(v < 0.1)$. The lift would then amount to $A = 0.789\, b^2\, V^2 \sin \alpha$, and the center of pressure rould lie at $s = 0.219\, b$.

III. RESULTS

By the methods described, the chordwise lift distribution was conputcd for elliptic spanwise distribution for five aspect ratios, namely, A = 1/4, 1/2, 1, 2, and 6, and graphically interpolated for arbitrary λ. The results are presented in figures 5 to 10 and are tabulated in the appendix. In figure 5 the coefficients for the circulation functions are plotted against A. A_1 Increases monotonically and for large aspect ratio approaches 2 as the asymptotic value. The absolute values of the other coefficients increase up to a maximum at about $A = \frac{1}{2}$, then drop rapidly to zero; A, and A, are always positive, and A_3 and A_4, negative. The smaller A is, the less rapidly do tbe A_ν converge, sothat to obtain the same accuracy as for large aspect ratios, a longer sorias of functions for the circulation must be assumed. The curve $A_1 \sqrt{b/\lambda}$ shows the increase in the circulation in tho neighborhood of the leading edge for constant span as a function of λ, since $\lim\limits_{x \to 0} \Upsilon(x) = A_1 \sqrt{\frac{b}{\lambda}}\, \frac{1}{\sqrt{x}}$; here, too, the maximum lies between $A = \frac{1}{2}$ and $\lambda = 1$. The valuo for A = 0 obviously is in orror, from which fact it may be seen that the given approximation for $\Upsilon(v)$ does not correctly represent the behavior in the neighborhood of the leading edge. If tho chord is held fixed and the span varied, A_1 itsolf gives the increase since $\lim\limits_{x \to 0} \Upsilon(x) = A_1 \sqrt{t/x}$.

On figures 6 and 7 the circulation distribution $\Upsilon(\xi)/V \sin \alpha$ and tho pressure difference between the lower and upper sidos of tho plate referrod to the dynamic pressure $\dfrac{p_u - p_o}{\frac{\rho}{2} V^2 \sin \alpha} = \dfrac{\pi}{2} \gamma(\xi)$ are plotted against the chord. In figure 6 the abscissa refers to the chord, and in figure 7, to the span. In tie first representation the limiting case $\lambda = 0$ coincide3 with the coordinate axes, so that tho lift distributions for any aspect ratio lie3 between tho axe3 and the limiting case $\lambda = \infty$. In tie second representation the limiting case $A = \infty$ coincides with axis of ordinates. This method of plotting is particularly susceptible to error3 in the circulation distribution and shows thnt at $A = \frac{1}{2}$, a small error is to be assumed through inaccuracy of one of the graphical quadratures or the approximated principal value. Similarly, the limiting case $\lambda = 0$ appears as only a very rough approximation since intersection of the curves with each other is very improbable. It is seen, however. that for very small aspect ratior a further increase in tio chord has only a small effect on the circulation distribution, either in the neighborhood of the leading edge or - owing to tie strong decrease - farther toward ths rear. This is seen especially clearly from the curve for $A/\rho\, b^2 V^2 \sin a$ (fig. 8), which shows how the total lift increases when tho span is held constant and the chord is varied.

On figures 9 and 10 the lift, drag, nnd moment coefficients are plotted as function3 of A. They are computed from the values of A_1, A_2, A_3, A_4 a3 follows: The lift according to tho Kutta-Joukowsky theorem is:

$$A = \rho V \int_{-b/2}^{+b/2} dy \int_0^t \gamma(x,y)\, dx = \frac{\pi^2}{8} \rho V b t \left(A_1 + \frac{1}{2} A_3 + \frac{1}{8} A_4\right)$$

from which $c_a = \dfrac{\pi^2}{4} \dfrac{1}{V} \left(A_1 + \dfrac{1}{2} A_2 + \dfrac{1}{8} A_4\right)$. Similarly, the moment about tho leading edge is

$$M = \rho V \int_{-b/2}^{+b/2} dy \int_0^t \gamma(x,y) x\, dx = \frac{\pi^2}{16} \rho V b t^2 \left(A_1 + A_2 + \frac{1}{4} A_3 + \frac{1}{4} A_4\right)$$

and $c_m = \dfrac{\pi^2}{16} \dfrac{1}{V} \left(A_1 + A_2 + \dfrac{1}{4} A_3 + \dfrac{1}{4} A_4\right)$ and finally the induced drag is

$$W_i = \rho \int_{-b/2}^{+b/2} dy \int_{0}^{t} \gamma(x,y)\ w(y)\ dx$$

where

$$dw(y) = \frac{1}{4\pi} \int_{-b/2}^{+b/2} \frac{\partial \gamma(x,\bar{y})}{\partial \bar{y}}\ dx\ \frac{d\bar{y}}{y-\bar{y}}$$

with

$$(x,y) = \frac{b}{2} \sqrt{1 - \left(\frac{y}{b/2}\right)^2}\ \sum_1^4 \nu\ A_\nu\ \gamma_\nu(x)$$

we have

$$\frac{dw(y)}{-ax} = - \frac{1}{2b} \sum_1^4 \nu\ A_\nu\ \gamma_\nu(x) = \text{const}$$

on account of tie elliptic spanwise distribution. so that $v_i = \frac{\pi}{4\lambda}\left(A_1 + \frac{1}{2} A_2 + \frac{1}{8} A_4\right)$ and $W_i = \frac{\pi^3}{32} \rho\ t^2 \left(A_1 + \frac{1}{2} A_2 + \frac{1}{8} A_4\right)^2$ and hence $c_{wi} = \frac{\pi^3}{16} \frac{1}{\lambda v^2} \left(A_1 + \frac{1}{2} A_2 + \frac{1}{8} A_4\right)^2$.

Hence in this case also, $\frac{c_a^2}{c_{wi}} = \pi \lambda$. On account of the elliptic circulation distribution. the factor $v = \pi/4$; hence multiplying by $\frac{4}{\pi} v(\lambda)$ to tako account of the v factor which incroasos with A, $(dc_a/dx)_0$ increases up to 2π as $\lambda \rightarrow \infty$. (The factor v was taken fron the dissertation by A. Pets.) The induced drag coefficient increases with increasing A, has a maximum at about $\lambda = 2$ and then. since the drag remains finite while the area becomes larger, drops to zero. The position of the center of pressure is obtained fron $s = (c_m/c_a)\ t$. This curve rapidly approaches tho asymptote. At $\lambda = 3$, the deviation fron the limiting value $s = 0.25\ t$ is only 6 percent (fig. 10).

The agreement of tie computation results by the vortex-filament method aith those by the vortex-sheet method is surprisinqlp good. Even with tso vortex filaments the deviations in the coefficients are small, whereas with four vortex filaments the deviations become large only for very deep plates with $\lambda \leq 1$. (For $\lambda = 1$: $\Delta c_a/c_a \approx -0.4$ percent, $\Delta c_m/c_m \approx +3.4$ percent; for $\lambda = 1/2$: $\Delta c_a/c_a \approx -1.6$ percent, $\Delta c_m/c_m \approx +90$ percent.) The individual lift portions contributed by the four strips of tho surface do not agree so well; for example:

	Four vortex filaments		Vortex sheet	
Lift from 0 - 3t/16:	0.711	b t V sin α	0.665	b t V sin α
3t/16 - 7t/16:	0,264	"	0.274	"
7t/16 - 11t/16:	0.141	"	c.144	"
11t/16 - t :	0.071	"	0.378	"

Tho circulation of the foremost vortex always cones out too high, and that of the other vortices too low. In obtaining the moment this error is partially compensated by the consideration that too large lever arms are used for the three rear vortex filaments, which do not lie at the centers of gravity of A_{II}, A_{III}, A_{IV}.

With both methods the assumption of elliptic distribution over the span - which assumption makes possible the solution of the integral equation in the case of the vortex sheet - should be the greatest source of error. For this reason, too low lift coefficicnts ore also obtained. The subsequent multiplication by the factor v does not appear to help sufficiently. According to the pressure distribution measurements of H. Winter which, however, are obtained for the square plate only, the distribution over the span at the loading edge is approximately elliptic, but farther toward the rear - almost up to the edge - It is constant, the edge disturbances which arise from the sharp edges of the investigated plate, however, not being taken into account.

APPENDIX

I. Several vortex filaments: Two vortex filaments with elliptic circulation distribution over the span at $x = t/8$ and $x = 5t/8$.

$\lambda = \frac{1}{2}$	$A_I = 0.3571\ btV \sin a$	$c_a = 0.770 \sin a$	$s = 0.161\ t$
	$A_{II} = 0.0277_2\ btV \sin a$	$c_m = 0.124$ sic a	
$\lambda = 1$	$A_I = 0.6132\ btV \sin a$	$c_a = 1.436$ sir a	$s = 0.198\ t$
	$A_{II} = 0.1050\ btV \sin a$	$c_m = 0.285 \sin a$	
A = 2	$A_I = 0.9410\ btV \sin a$	$c_a = 2.369 \sin a$	$s = 0.228\ t$
	$A_{II} = 0.2434\ btV \sin a$	$c_m = 0.540 \sin a$	
A = 6	$A_I = 1.4280\ btV \sin a$	$c_a = 3.767 \sin a$	$s = 0.246\ t$
	$A_{II} = 0.4557\ btV \sin a$	$c_m = 0.923 \sin a$	

Tour vortex filaments with elliptic circulation distribution over the span at x = t/16, 5t/16, 9t/16, and 13t/16.

		Vortex shoot	
$\lambda = \frac{1}{2}$	$A_I = 0.5080\ btV$ sir a	(0.2979)	$c_a = 0.772 \sin a$
	$A_{II} = 0.0557$ "	(0.0724)	$c_w = 0.379 \sin^2 \alpha$
	$A_{III} = 0.0165$ "	(0.0154)	$c_m = 0.101 \sin a$
	$A_{IV} = 0.0058$ "	(0.0067)	$s = 0.131\ t$
A = 1	$A_I = 0.4876\ btV \sin a$	(0.4838)	$c_a = 1.441 \sin a$
	$A_{II} = 0.1433$ "	(0.1470)	$c_w = 0.661 \sin^2 \alpha$
	$A_{III} = 0.0623$ "	(0.0627)	$c_m = 0.265 \sin a$
	$A_{IV} = 0.0273$ "	(0.0299)	s 40.184 t
$\lambda = 2$	$A_I = 0.7111\ btV \sin a$	(0.6852)	$c_a = 2.374$ sir a
	AI I $= 0.2644$ "	(0.2739)	$c_w = 0.897 \sin^2 \alpha$
	$A_{III} = 0.1405$ "	(0.1442)	$c_m = 0.528 \sin a$
	$A_{IV} = 0.0710$ "	(0.0782)	$s = 0.222\ t$

Vortex sheet

$\lambda = 6$ $A_I = 1.0480\ btV \sin\alpha$ (1.0068) $c_a = 3.770 \sin\alpha$

$A_{II} = 0.4385$ " (0.4425) $c_w = 0.754 \sin^2\alpha$

$A_{III} = 0.2576$ " (0.2525) $c_n = 0.923 \sin\alpha$

$A_{IV} = 0.1406$ " (0.1465) $a = 0.245\ t$

II Vortex surface: Examples for $\lambda < 2$

$\lambda = \frac{1}{4}$ $J_1 = 6.3138$ $J_2 = 0.6725$ $J_3 = -0.7284$ $J_4 = 0.5566$

$J_5 = 8.2523$ $J_6 = 2.4674$ $J_7 = -1.1575$ $J_8 = 0.6169$

$J_9 = 9.4427$ $J_{10} = 4.0623$ $J_{11} = -0.7284$ $J_{12} = 0.6772$

$J_{13} = 10.0592$ $J_{14} = 5.2067$ $J_{15} = 0.2581$ $J_{16} = 1.4606$

$A_1 = 0.3435\ v \sin\alpha$ $c_a = 0.3839 \sin\alpha$ $A_1/\sqrt{\lambda} = 0.687\ V \sin\alpha$

$A_2 = -0.3307$ " $c_w = 0.1026 \sin^2\alpha$ total lift:

$A_3 = 0.2994$ " $c_n = 0.0287 \sin\alpha$ $A = 0.7778\ \rho\ b^2\ V$

$A_4 = -0.1542$ " $s = 0.0739\ t$

$\lambda = \frac{1}{2}$ $J_1 = 6.8189$ $J_2 = 0.6345$ $J_3 = -1.0013$ $J_4 = 0.6956$

$J_5 = 8.6508$ $J_6 = 2.4574$ $J_7 = -1.5644$ $J_8 = 0.6159$

$J_9 = 9.7617$ $J_{10} = 4.2703$ $J_{11} = -1.0013$ $J_{12} = 0.5381$

$J_{13} = 10.3816$ $J_{14} = 5.8817$ $J_{15} = 0.5725$ J_{16} -1.7546

$A_1 = 0.5793\ v \sin\alpha$ $c_a = 0.7846 \sin\alpha$ $A_1/\sqrt{\lambda} = 0.819\ V \sin\alpha$

$A_2 = -0.4802$ " $c_w = 0.3919 \sin^2\alpha$ $A = 0.7846\ \rho\ b^2\ V^2$

$A_3 = 0.3710$ " $c_n = 0.0922 \sin\alpha$

$A_4 = -0.1697$ " $s = 0.1175\ t$

$\lambda = 1$ $J_1 = 8.1785$ $J_2 = 0.1265$ $J_3 = -1.1401$ $J_4 = 0.8755$

$J_5 = 9.6696$ $J_6 = 2.4674$ $J_7 = -1.9864$ J_8 -0.6169

$J_9 = 10.6560$ $J_{10} = 4.8083$ $J_{11} = -1.1401$ $J_{12} = 0.3583$

$J_{13} = 11.3206$ $J_{14} = 7.0036$ $J_{15} = 1.3577$ $J_{16} = 2.4958$

$A_1 = 0.8182\ V \sin a$ $c_a = 1.4466 \sin a$ $A_1/\sqrt{\lambda} = 0.818\ V \sin \alpha$

$A_2 = -0.4424$ " $c_w = 0.6662 \sin^2 \alpha$ $A = 0.7233\ \rho\ b^2\ V^2$

$A_3 = 0.2440$ " $c_m = 0.2563 \sin \alpha$

$A_4 = -0.0852$ " $s = 0.1771\ t$

Examples for $\lambda \geq 2$

$\lambda = 2$ $F_1 = 3.6683\pi$ $F_2 = -0.4078\pi$ $F_3 = -0.5899\pi$ $F_4 = 0.4204\pi$,

$F_5 = 3.9707\pi$ $F_6 = 0.7854\pi$ $F_7 = -1.1236\pi$ $F_8 = 0.1964\pi$

$F_9 = 4.1905\pi$ $F_{10} = 1.9786\pi$ $F_{11} = -0.5898\pi$ $F_{12} = -0.0277\sim$

$F_{13} = 4.3408\pi$ $F_{14} = 3.1208\pi$ $F_{15} = 0.9732\pi$ $F_{16} = 1.2815\pi$

$A_1 = 1.0790\ V \sin \alpha$ $c_a = 2.3629 \sin a$ $A_1/\sqrt{\lambda} = 0.763\ V \sin a$

$A_2 = -0.2389$ " $c_w = 0.8886 \sin^2 \alpha$ $A = 0.5937\ \rho\ b^2\ V^2$

$A_3 = 0.0834$ " $c_m = 0.5288 \sin a$

$A_4 = -0.0149$ " $s = 0.2238\ t$

$\lambda = 6$ $F_1 = 7.6668\pi$ $F_2 = -2.3693\pi$ $F_3 = -1.5710\pi$ $F_4 = 0.9304\pi$

$F_5 = 7.8924\pi$ $F_6 = 0.7854\pi$ $F_7 = -3.1037\pi$ $F_8 = 0.1964\pi$

$F_9 = 8.0425\pi$ $F_{10} = 3.9401\pi$ $F_{11} = -1.5710\pi$ $F_{12} = -0.5377\pi$

$F_{13} = 8.1334\pi$ $F_{14} = 7.0466\pi$ $F_{15} = 2.9898\pi$ $F_{16} = 3.2617\pi$

$A_1 = 1.5449\ V \sin\alpha$ $\quad c_a = 3.6946 \sin\alpha$ $\quad A_1/\sqrt{\lambda} = 0.631\ V \sin\alpha$

$A_2 = -0.09344$ " $\quad c_w = 0.7242 \sin^2\alpha$ $\quad A = 0.3079\ \rho\ b^2\ V^2$

$A_3 = 0.03808$ " $\quad c_m = 0.9002 \sin\alpha$

$A_4 = -0.00630$ " $\quad s = 0.2437 \cdot t$

$\lambda = \infty \quad A_1 = 2V \sin\alpha \quad c_a = \frac{\pi^2}{2} \sin\alpha$ instead of $c_a = 2\pi \sin\alpha$

$A_2 = A_3 = A_4 = 0 \quad c_m = \frac{\pi^2}{8} \sin\alpha$ " " $c_m = \frac{\pi}{2} \sin\alpha$

$s = 0.25\ t$

Translation by S. Reiss,
National Advisory Committee
for Aeronautics.

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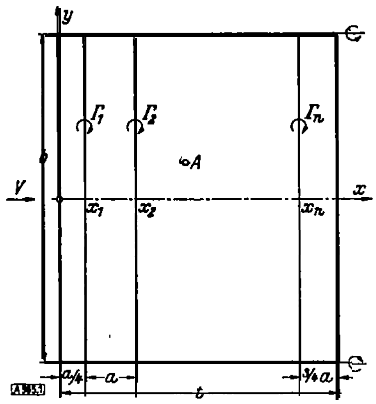

Figure 1.

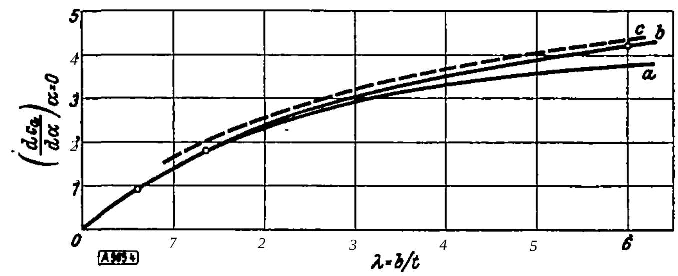

Figure 4.- Lift **distribution as** a function of λ for a vortex filament, **a,** $\Gamma(\varphi)=\Gamma \sin\varphi$; b, $\Gamma(\varphi)=\Gamma \sin\varphi(1+a_1 \sin\varphi)$; c, **test** result8 of Winter.

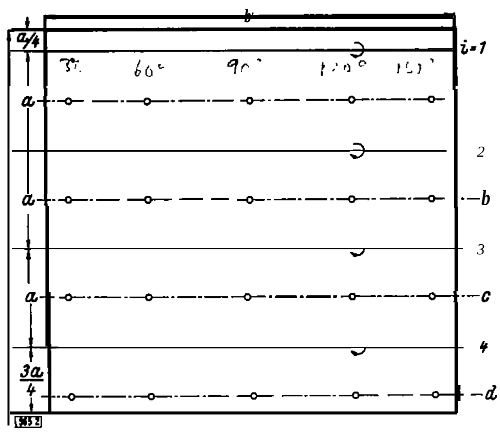

Figure 2.

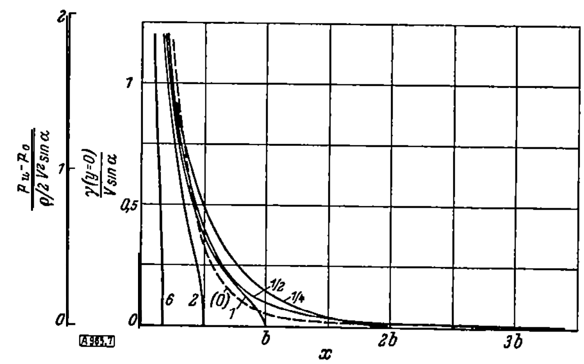

Figure **7.-** Circulation dietribution over the chord; b= conet.

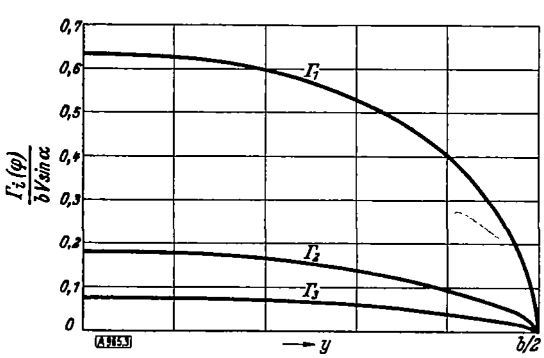

Figure **3.-** Circulation dietribution of the forward three vortex filaments at **x**=**t/16, 5t/16, 9t/16;**λ= 1.

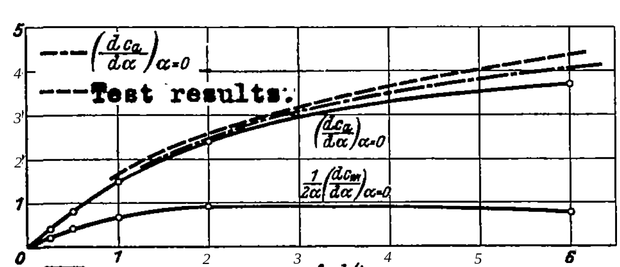

Figure 9.- Lift **and** drag **coefficients as** functions ofh.

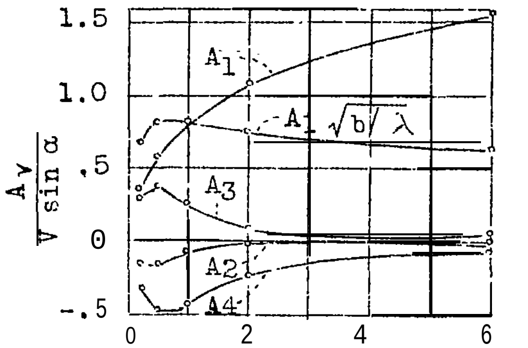

Figure 5.- Coefficients cff circulation functions for various aspect ratios.

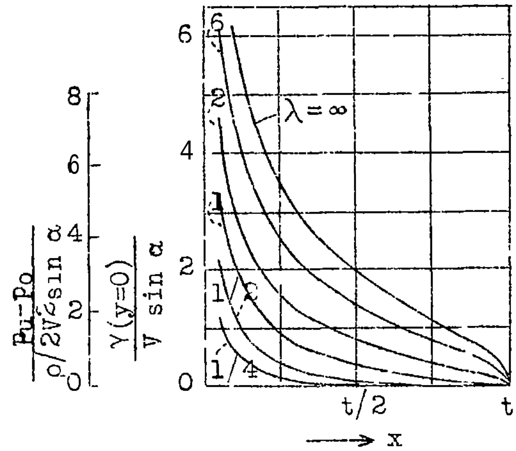

Figure 6.- Circulation distribution over the chord; t = const,

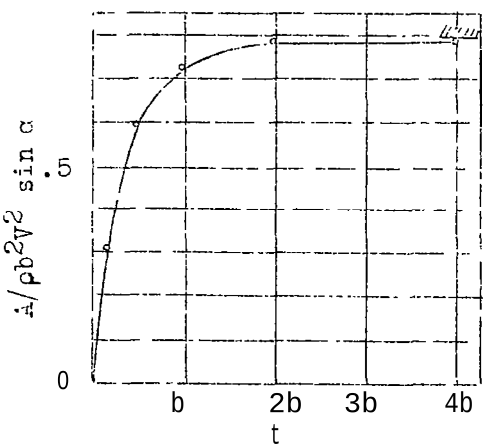

Figure 8.- Total lift as a function of the chord.

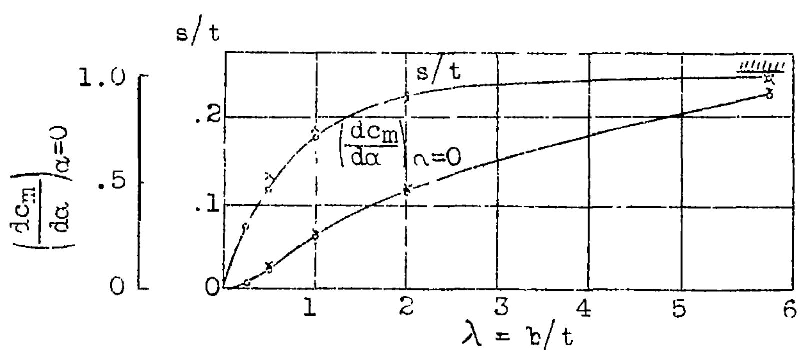

Figure 10.- Moment coefficient and distance of center of pressure from leading edge as functions of λ. (points x computed with four vortex lines).